

THE *Hinode* (SOLAR-B) MISSION: AN OVERVIEW

T. KOSUGI, K. MATSUZAKI, T. SAKAO, T. SHIMIZU, Y. SONE, S. TACHIKAWA,
K. MINESUGI, A. OHNISHI, T. YAMADA,

*Institute of Space and Astronautical Science, Japan Aerospace Exploration
Agency, Sagami-hara, Kanagawa 229-8510, Japan*

S. TSUNETA, H. HARA, K. ICHIMOTO, Y. SUEMATSU, M. SHIMOJO, T.
WATANABE,

National Astronomical Observatory of Japan, Mitaka, Tokyo 181-8588, Japan

S. SHIMADA,

*Kamakura Works, Mitsubishi Electric Corp., Kamakura, Kanagawa 247-8520,
Japan*

J. M. DAVIS, L. D. HILL, J. K. OWENS,

*Space Science Office, VP62, NASA Marshall Space Flight Center, Huntsville, AL
35812, U.S.A.*

A. M. TITLE,

*Lockheed Martin Solar and Astrophysics Laboratory, B/252, 3251 Hanover Street,
Palo Alto, CA 94304, U.S.A.*

J. L. CULHANE, L. K. HARRA,

*UCL Mullard Space Science Laboratory, Holmbury St. Mary, Dorking, Surrey
RH5 6NT, UK*

G. A. DOSCHEK,

*E. O. Hulburt Center for Space Research, Code 7670, Naval Research Laboratory,
Washington, DC 20375-5352, U.S.A.*

and

L. GOLUB

Smithsonian Astrophysical Observatory, Cambridge, MA 02138, U.S.A.

Abstract: The *Hinode* satellite (formerly Solar-B) of the Japan Aerospace Exploration Agency's Institute of Space and Astronautical Science (ISAS/JAXA) was successfully launched in September 2006. As the successor to the *Yohkoh* mission, it aims to understand how magnetic energy is transferred from the photosphere to the upper atmospheres and resulting in explosive energy releases. *Hinode* is an observatory style mission, with all the instruments being designed and built to work together to address the science aims. There are three instruments onboard: the Solar Optical Telescope (SOT), the EUV Imaging Spectrometer (EIS), and the X-ray Telescope (XRT). This paper overviews the mission, including the satellite, the scientific payload and operations. It will conclude with discussions on how the international science community can participate in the analysis of the mission data.

Keywords: Solar-B · *Hinode* · solar magnetic fields – solar corona – solar variability

1. Introduction

The *Hinode* spacecraft (formerly Solar-B) of the Institute of Space and Astronautical Science, Japan Aerospace Exploration Agency (ISAS/JAXA) was launched on September 22, 2006 at 21:36 GMT, aboard the seventh in JAXA's series of M-V rockets. The principal scientific goals of the *Hinode* mission are:

- 1) To understand the processes of magnetic field generation and transport including the magnetic modulation of the Sun's luminosity
- 2) To investigate the processes responsible for energy transfer from the photosphere to the corona and that are responsible for the heating and structuring of the chromosphere and the corona.
- 3) To determine the mechanisms responsible for eruptive phenomena, such as flares and coronal mass ejections, and understand these phenomena in the context of the space weather of the Sun-Earth System.

The mission is the follow-on to *Yohkoh* an ISAS mission, with significant NASA and UK participation that was launched in 1991 (Ogawara et al. 1991), and continued taking observations for nearly a solar cycle. *Yohkoh* demonstrated that the high temperature corona is highly structured and dynamic and that rapid heating and mass acceleration are common phenomena (Acton et al 1992). *Yohkoh* was launched shortly after the maximum of solar cycle 22, which was ideal period for studying large solar flares. The subsequent observations provided considerable evidence to support magnetic reconnection as the driver for energy release in flares. Hard X-ray 'above the loop top' sources were found in compact flares (e.g. Masuda et al. 1994) and also in long duration flares (e.g. Harra et al. 1998). In soft X-rays the flaring loops often took on the appearance of cusps which is to be expected from the standard model where the reconnection occurs high in the corona (e.g. Tsuneta 1996, Canfield et al 1999, Sterling et al 2000). The edges of the loops were also found to be hotter as expected if the outer edges are the last to be heated from reconnection. As expected from the reconnection, plasma ejections from flaring site has been found on many occasions (e.g. Shibata et al. 1995). On smaller scales, many jets were found in soft X-rays, which are interpreted as reconnection occurring through the interaction of emerging flux and already existing magnetic field (Shimojo et al, 1996). Many small-scale flares were observed in active region loops (e.g. Shimizu 1995, Shimizu et al. 2003), and in bright points (Priest et al. 1994). On the more global scale, dramatic coronal waves were observed (e.g. Hudson et al. 2003) and trans-equatorial loops were found to erupt (e.g. Khan and Hudson 2000) followed by coronal mass ejections or flares (Harra et al. 2003).

Hinode is designed to address the fundamental question of how magnetic fields interact with the ionized atmosphere to produce solar variability. Measuring the properties of the Sun's magnetic field is the fundamental observational goal of *Hinode* and is its essential difference from previous solar missions. The three instruments were selected in order to observe the response of the chromosphere and corona to changes in the photospheric magnetic field. To achieve this end *Hinode* makes quantitative measurements of all three components of vector magnetic fields. This allows the free energy of the magnetic field to be calculated, which through the action of electric currents powers solar activities. The components of the magnetic field are difficult to resolve especially from the ground where seeing effects degrade spatial resolution. The major scientific instrument on *Hinode*, the solar optical telescope (SOT) makes these observations from space. The response of the solar atmosphere to field changes is measured by an EUV imaging spectrometer (EIS), and an X-ray telescope (XRT).

Based upon this scientific motivation, *Hinode* was planned and constructed as an international collaborative project including institutions in Japan, in the United States, and in the United Kingdom. ISAS/JAXA has responsibility for the design, development, test and integration of the *Hinode* spacecraft with National Astronomical Observatory of Japan as a domestic partner and the Mitsubishi Electric Corporation as a leading contractor. The participating institutes and

their responsibilities are shown in Table I. The *Hinode* spacecraft was called by its development name *Solar-B* and the name *Hinode* was given after successful launch according to the Japanese satellite tradition. *Hinode* is a Japanese word meaning sunrise.

In the present paper, we will give an overview of the *Hinode* mission from the viewpoints of the scientific instruments in Section 2, the spacecraft design in Section 3, and the flight operations in Section 4. The scientific objectives will be briefly discussed in Section 5.

Table I
The Solar-B mission

Mission objective	Investigation of magnetic activities of the Sun including its generation, energy transfer and release of the magnetic energy
Launch	September 22, 2006, 21:36UTC
Mission Life	≥ 3 year
Organization	
Project manager	T. Kosugi ¹ (ISAS/JAXA)
Co-manager	S. Tsuneta (NAOJ)
Project Scientists	T. Sakurai (NAOJ), K. Shibata (Kyoto University), J. M. Davis (MSFC), and L. K. Harra (MSSL) ²
Principal investigators	
Solar Optical Telescope (SOT) :	
S. Tsuneta (NAOJ), and A. M. Title (LMATC) ³	
EUV Imaging Spectrometer (EIS) :	
J. L. Culhane (MSSL) ⁴ , and G. A. Doschek (NRL), and T. Watanabe (NAOJ)	
X-ray Telescope (XRT) :	
L. Golub (SAO) ⁵ , and K. Shibasaki (NAOJ)	
Responsible Institute	
The Japan Aerospace Exploration Agency's Institute of Space and Astronautical Science (ISAS/JAXA) : Overall mission including the launch vehicle	
National Astronomical Observatory of Japan (NAOJ) :	
Three scientific instruments and support for spacecraft development	
The National Aeronautics and Space Administration (NASA) :	
Three scientific instruments	
The Particle Physics and Astronomy Research Council (PPARC) ⁶ : EIS	
European Space Agency (ESA) : Ground station support	
Major participating institutions	
Scientific instruments are built by collaborative efforts of the following institutes	
SOT : NAOJ, Lockheed Martin Solar and Astrophysics Laboratory (LMSAL), High Attitude Observatory (HAO), ISAS/JAXA, NASA	
EIS : Mullard Space Science Lab. (MSSL), U.S. Naval Research Laboratory (NRL), NAOJ, ISAS/JAXA, Rutherford Appleton Laboratory (RAL), Birmingham University, The University of Oslo	
XRT : Smithsonian Astrophysical Observatory (SAO), ISAS/JAXA, NAOJ, NASA	

¹ I. Nakatani as project manager and T. Sakao and T. Shimizu as deputy project managers after T. Kosugi passed away in November 2006.

² Succeeded by D.R. William in 2006.

³ Succeeded by T. D. Tarbell in 2004.

⁴ Succeeded by L.K. Harra in 2006.

⁵ Succeeded by E. E. Deluca in 2005.

2. Scientific Instruments

The scientific payload consists of three instruments: the Solar Optical Telescope (SOT), the EUV Imaging Spectrometer (EIS), and the X-ray Telescope (XRT). Each instrument is a result of the combined talents of all the members of the international team. Full technical details of each instrument are described in the separate papers in this special issue. This paper provides a brief summary of each instrument with their main characteristics summarized in Table II. The instruments usually work together as an 'observatory' studying the same target at which the spacecraft is pointed. Optionally, the EIS has the ability to offset its own pointing and the XRT, having larger field of view than the others, has the ability to observe its own region of interest.

Table II

Solar-B Scientific Instruments

(a) Properties of the Telescopes

Solar Optical Telescope (SOT)	
<i>Optical Telescope Assembly (OTA)</i>	
Optics	Aplanatic Gregorian with aperture of 50cm
<i>Focal Plane Package (FPP)</i>	
Wavelength and lines	<u>Broadband Filter Instrument (BFI)</u> CN I(3883.0), Ca II H (3968.5), CH I (4305.0) Blue (4504.5), Green (5550.5), Red (6684.0) <u>Narrowband Filter Instrument (NFI)</u> Mg Ib (5172.7), Fe I(5250.2, 5247.1, 5250.6) , Fe I(5576.1), Ns I (5895.9), Fe I (6302.5, 6301.5), H I (6562.8) <u>Spectro Polarimeter (SP)</u> Fe I (6302.5, 6301.5)
Resolution of	longitudinal : 1-5G
Magnetic Field	transverse : 30-50G
Typical Time Cadence	Ranges from tens of seconds for photospheric images and vector magnetographs in particular lines to ~ 1hr for the full Stokes profiles.
EUV Imaging Spectrometer (EIS)	
Optics	Offset paraboloid with multi layer coating mirror and concave grating with aperture of 15cm
Wavelength	170Å-210Å with resolution of ~4000 250Å-290Å with resolution of ~4600
Velocity Resolution	3km/s for Doppler velocity, 20km/s for line width
Exposure Time	ms in flares, tens of seconds in active regions
X-Ray Telescope (XRT)	
Optics	Modified Wolter type I grazing incident mirror and co-aligned optical telescope
Wavelength	X-ray: 2Å-200Å Optical: G-band (4305Å)
Temperature	$\Delta \text{Log } T : 0.2^1$
Discrimination	
Exposure Time	4ms-10s

¹ In the case of isothermal plasma

(b) Properties of the Focal Plane Detectors

Instruments	F.O.V. EW x NS (Slit/Slot)	Pixel Size
SOT		
NFI ² :	328"x164"	0.08"
BFI ² :	218"x109"	0.053"
SP:	320"x164" (0.16")	0.16" x 21.5mÅ
EIS	590"x512" (1", 2", 40", 266")	1.0" x 0.0223Å
XRT	2048"x2048"	1.0"

² NFI and BFI share a CCD

2.1. Solar Optical Telescope (SOT)

The SOT is the largest, solar, optical telescope flown in space (Tsuneta et al. 2007). The SOT consists of the Optical Telescope Assembly (Suematsu et al. 2007) and its Focal Plane Package (Tarbell et al. 2007). The OTA is a 50cm clear aperture, aplanatic Gregorian, F9 design telescope. The OTA is diffraction limited (0.2"-0.3") between 3880-6700Å. The primary mirror is fabricated out of ULE and supported by invar/titanium structures retaining thermal stability. Field stops and heat rejection mirrors are located at the focus of the primary mirror and at the Gregorian focus. The secondary field stop limits the field of view to 361" x 197". The OTA holds the collimating lens unit (CLU), the polarization modulator (PMU), and a tip-tilt mirror (CTM) behind the primary mirror. The PMU is a continuously rotating waveplate optimized for linear and circular polarization at 5173Å and 6302Å. The SOT is well designed and calibrated for performing the measurements of polarization with high accuracy (Ichimoto et al. 2007). With the CLU and the tip-tilt mirror, the OTA provides a pointing-stabilized parallel beam to the FPP. The FPP has four optical paths: the Narrowband Filter Instrument (NFI), Broadband Filter Instrument (BFI), Spectro-polarimeter (SP) and Correlation Tracker (CT). The BFI and the NFI share a CCD detector and constitute the Filtergraph (FG). The SP and the CT have their own CCD detectors. The NFI uses a tunable Lyot, birefringent, filter to record filtergrams, Dopplergrams, and longitudinal and vector magnetograms across the spectral range from 5170-6570 Å. The BFI has interference filters to image the photosphere and low chromosphere and to make blue, green and red continuum measurements for irradiance studies. The SP is an off-axis Littrow echelle spectrograph that records dual-line, dual beam, polarization spectra of the FeI 6302.5Å and FeI 6301.5Å spectral lines for high precision Stokes polarimetry. The CT is the high speed CCD camera to sense jitter of solar features on the focal plane, which jitter signal is fed to the closed-loop control of the tip-tilt mirror (Shimizu et al. 2007). This image stabilization system prevents the spacecraft jitter from affecting the resolution of the images. The image stabilization system achieves a stability of 0.007" (3σ) below the cross over frequency of 14Hz. Time-line sequence of the data acquisitions by the SOT is controlled according to observation tables (one for FG and the other for SP) on the Mission Data Processor (MDP).

2.2. EUV Imaging Spectrometer (EIS)

The EUV Imaging Spectrometer (EIS, Culhane et al., 2007) is an imaging spectrometer designed to observe plasmas in the temperature range from 0.1MK, the upper transition, to 10MK, the lower corona. The EIS is an off-axis paraboloid telescope with a focal length of 1.9m and a mirror diameter of 15cm. The angular resolution of the optics is 2". The total length of the instrument is 3m. The primary mirror has a mechanism that can offset the field of view of the EIS in the E-W direction relative to the spacecraft pointing. The mirror

illuminates various slits that are placed at the focus of two multilayer coated, toroidal gratings that disperse the spectrum onto two back-side illuminated CCD detectors. The detectors cover the wavelength ranges of 170-210Å and 250-290 Å with spectral resolution of $R \sim 4000$. Four slit or slot widths are available - 1" slit, 2" slit, 40" slot and 266" slot. High spectral resolution images can be obtained by rastering with the slit. The slots provide "overlappograms" of the transition region and corona at high cadence. The EIS instrument provides factors of three improvement in spatial and spectral resolution and sensitivity over CDS (Coronal Diagnostic Spectrometer) aboard the SOHO spacecraft. The velocity resolution is 3km/s for Doppler velocities and 20km/s for line widths. With the higher sensitivity and higher telemetry rate of the spacecraft, the EIS can achieve a time cadence of 0.5s in flares and ~ 10 s in active regions. The control system is designed to optimize the use of the telemetry allocation. It provides the flexibility to select the mix of spectral lines, image regions, and time cadence of an observation to match specific scientific objectives. A dedicated processor within EIS provides the control function and can operate autonomously to switch observations in response to notification of a flare by the XRT or detection of a flare or a bright point by the EIS processor itself.

2.3. X-ray Telescope (XRT)

The X-ray telescope (XRT) is a grazing incidence telescope of a Wolter I design made from Zerodur (Golub et al. 2007). The mirror has a 30cm aperture and a 2.7m focal length. The surface figure is a modified paraboloid-hyperboloid whose surfaces are optimized to minimize the blur circle radius at large angles. The reflecting surfaces are uncoated and, together with improved entrance filters that reject the Sun's visible light, provide a lower energy cutoff of X-ray than SXT aboard *Yohkoh*. In front of the focal plane, there are two filter wheels containing a total of nine X-ray analysis filters, which pass wavelength bands with different lower cutoff energy. Because of the lower cutoff energy, the XRT can observe plasmas with temperatures as low as 1×10^6 K in the lower corona. The brightness ratio between images taken through two different filters provides a measure of the temperature of the plasma when the observed plasma can be assumed to be isothermal. For flare studies the filter ratio method is capable of measuring temperatures as high as $\sim 30 \times 10^6$ K. In addition to the X-ray optics, the XRT is equipped with visible light optics, to be used with a G-band filter, for the purpose of co-alignment of XRT and SOT images. The X-ray and visible light optics share the focal plane where a back-side illuminated CCD is located. The CCD has a pixel size of 1 arcsec and the field of view is 34×34 arcmin² which covers the whole solar disk when the spacecraft is pointed at sun-center. The CCD camera is equipped with an on-orbit focus adjustment mechanism (Kano et al. 2007). The camera is launched out of focus and in addition to moving the camera to the best on-orbit focus it can also be used to optimize the across the field of view to compensate for field curvature. For example, for the highest resolution observations an on-axis focus provides an angular resolution of ~ 1 arcsec within a radius of ~ 7 arcmin. For the best resolution across the field of view the focus can be moved forward to provide an angular resolution of ≤ 3 arcsec within a radius of ~ 17 arcmin. The camera, its shutter for exposure control, and the filter wheel are controlled according to observation sequence defined as observation table in the MDP. To optimize the use of the telemetry allocation the field of view, filter sequence and time cadence can be adjusted to match each scientific objective. MDP also has various functions for enhancing XRT observations, including automatic region selection, automatic exposure duration control, flare detection, and memory buffer for storing high-cadence images taken in pre-flare phase.

3. The Spacecraft

3.1. General

The *Hinode* spacecraft was launched from the Uchinoura Space Center at latitude 31 N, longitude 131 E, by the seventh, and last, a M-V launch vehicle into an elliptical polar orbit with a perigee of ~280km and apogee of ~686km. In the succeeding phase, the *Hinode* spacecraft boosts its perigee and controls the plane of the orbit with its own thrusters to acquire a circular, sun-synchronous, polar orbit of about 680 km altitude, 98.1 deg inclination, and 98 min period. With this orbit, *Hinode* can observe the sun continuously for a duration of 9 month each year.

The major parameters of the spacecraft are summarized in Table III. The spacecraft, schematically shown in Figure 1, has dimensions of approximately 4000 x 1600 x 1600 mm with two external solar panels (4300 x 1100 mm each) outside and weights about 900 kg. Three telescopes are aligned in the Z axis of the spacecraft and supported by an optical bench unit (OBU). The OBU is a cylinder made up of composite material that supports the OTA internally. The FPP, EIS and XRT are kinematically mounted on the outside of the OBU with 6 mounting legs, which constrain the degrees of freedom of the rigid body. The OBU also holds a tower to whose upper surface the sun sensors are attached. The electronics units are located in the bus box attached to the bottom of the OBU. The solar cell panels are designed to supply about 1100 W during spacecraft day. Excess power is either stored into NiCd batteries to supply the power required during spacecraft night or is consumed by a shunt regulator.

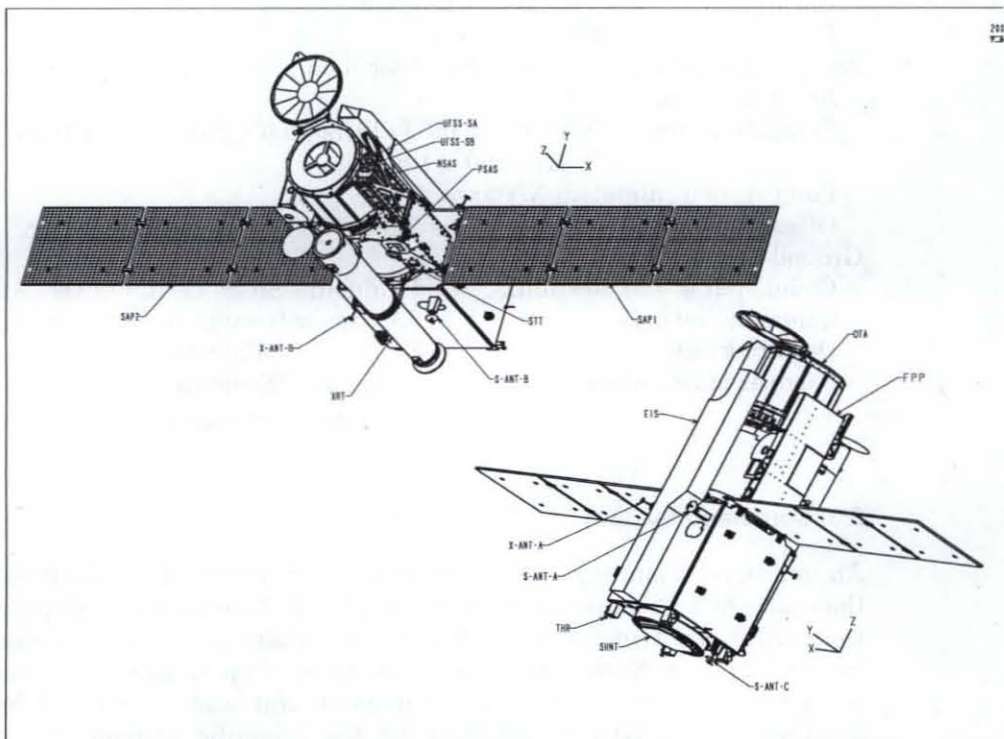


Figure 1. The Solar-B Spacecraft and its scientific instruments.

3.2 Attitude and Orbit Control

The *Hinode* spacecraft is stabilized by the attitude and orbit control system (AOCS) in three axes with its Z-axis pointed to the Sun. The Y-axis is directed toward the solar north. As a baseline, the spacecraft tracks a region on the solar surface by correcting for solar rotation. For each tracked target, the angular velocity around the rotation axis of the Sun can be specified. The another mode is the spacecraft pointing to a fixed position on the solar disk. In either case, stability of the Z-axis is 0.3" (3σ) per 10 sec and 1" per min.

The AOCS uses momentum wheels, and magnetic torquers, as the actuators for attitude control and thrusters for orbital control. The attitude sensors consist of two types of sun sensors including two fine sun sensors (UFSS), a star tracker, and geomagnetic sensors are available for determining spacecraft pointing relative to the direction of the Sun and to the ecliptic plane while an inertial reference unit comprising four gyros detects changes of attitude with time. Signals from two UFSS sun sensors with random noise level of 0.3" (3σ) can be used to remove the jitter of the satellite Z-axis pointing from the time series of data.

Table III

Major Parameter of Solar-B	
Size:	4000 x 1600 x 1600 mm
Weight:	900 kg (Wet), 770 kg (Dry)
Power:	1100 W
Data rate:	up to 2 Mbps (science data), and 32kbps (housekeeping)
Data recorder:	8 Gbits
Telemetry rate:	32 kbps (S-band), 4 Mbps (X-band)
Orbit:	
Altitude:	680 km attitude (circular, Sun-synchronous, polar)
Inclination:	98.1 deg inclination
Period:	98 min period
Attitude control (requirement):	3-axis stabilized
Absolute pointing	20"
Stability around	X/Y-axes: 0.06" (>20Hz), 0.6" / 2sec, 4.5" / 1 hour Z-axis: 200" / 1 hour
Pointing determination	X/Y-axes: 0.1"
Offset pointing	Up to 1178" from the Sun center
Ground Stations	
Commanding and downlink	Uchinoura Space Center (131E, 31N)
Commanding only	JAXA new Ground Network stations
Downlink only	Svalbard (15E, 78N)
Number of Downlinks	15 per day (Svalbard) 4 per day (Uchinoura)

3.3. Command System

An uplink commanding system controls the operation of all the instruments on the spacecraft. Commands are sent from the Uchinoura Space Center as well as from JAXA new Ground Network antennas. There are about 3 contacts in a day for commanding. Each contact has a duration of up to 10 min. Commands from the ground are received by the command unit and distributed by the data handling unit (DHU). Commands for the scientific instruments are further relayed by the mission data processor (MDP). The DHU can coordinate commands into sequences called 'organized commands' (OG's). The DHU can store up to 512 sets of OG's, each being a set of up to 8 commands. First, an OG

can be launched by a 'Real time OG execute command' from the ground. Second, a series of OG's can be dispatched sequentially with specified time intervals by the DHU itself. Such a series is called an 'operation program (OP)'. The OP can contain up to 4096 OG references. The OP is initiated by an 'OP start command'. The OP can last for up to about 10 days, so that the operation can be programmed before hand. In addition to the OG, the DHU can store sequences of commands to be executed during spacecraft emergencies. These are triggered by the AOCS or autonomous detection of an emergency by the DHU. The latter case includes failure modes of the battery system.

3.4. Onboard Data Processing

Observations of the three scientific instruments are governed by the mission data processor (MDP). Figure 2 is a schematic representation of the onboard observation control system. In the case of the FG, SP, and XRT, the MDP controls the observations. The controls are implemented using observation tables that make use of programs which have a nested loop-call structure. The EIS instrument's observing sequences are controlled by its own processor. In addition to normal observations, the scientific instruments have the capability to switch autonomous observations when notified by the onboard system of a flare. The MDP continuously analyzes XRT images for large intensity increases indicative of a flare. If a flare is found a flare flag is issued that allows the instruments to terminate their current sequence and switch to a flare observation program. The observation table for flare studies has the same structure as those for normal observations.

The scientific data from the instruments are compressed in the MDP before being stored in the data recorder. Memory space is divided between the SOT, XRT and EIS in the ratio of 70:15:15. The MDP has a compression speed of 832 Kpixel/sec for SOT, 256 Kpixel/sec for XRT, and 128 Kpixel/sec for EIS which matches the data acquisition rate and storage capacity for each instrument.

Two types of compression are performed sequentially. The first is pixel by pixel bit compression followed by image compression. The pixel by pixel bit compression is based on look up tables and implemented by hardware. In the table a smooth function composed of a linear and quadratic components part can be registered. The image compression is either a lossless compression using a DPCM (Differential Pulse Code Modulation) algorithm or a lossy compression using a DCT (Discrete Cosine Transform) algorithm (JPEG). These schemes are implemented by an application specific integrated circuit (ASIC). Parameters for compressions, which affect the compression ratio and data quality, can be optimized on orbit.

The MDP can output compressed data from the SOT, XRT, and EIS at rates of up to 1.3 Mbps, 262 kbps, and 262kbps, respectively. The actual data rates from the telescopes are determined by the observation tables and compression efficiency. During the preparation of the observation table care has to be taken to ensure that they are consistent with the duration of downlink contacts. The tables should be implemented in scientific operation described in Section 4.2.

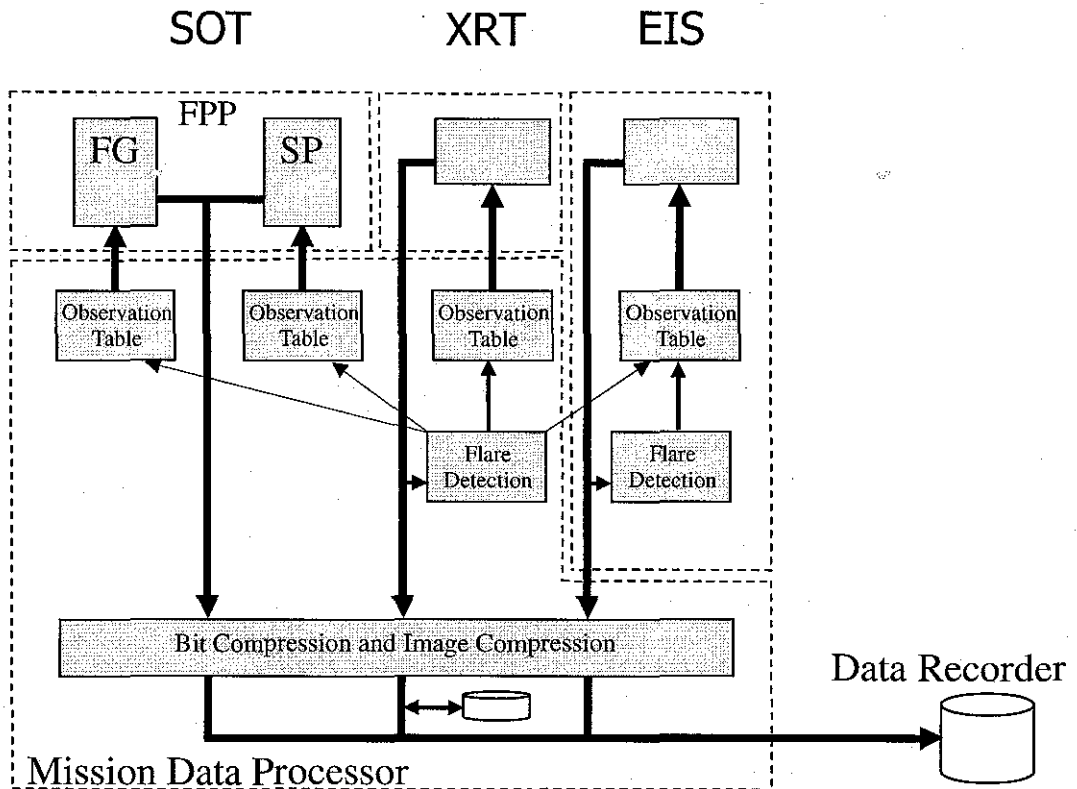


Figure 2. Functional Block Diagram of Onboard Observation Control System

3.5. Data Recording

Telemetry from the spacecraft follows the specification of packet recommended by the Consultative Committee for Space Data Systems (CCSDS). Telemetry packets from the scientific instruments are edited by the MDP while the housekeeping and spacecraft data are edited by the DHU.

The *Hinode* spacecraft has limited duration, ground-station contacts. Telemetry packets that cannot be down linked during a particular ground-station contact remain stored in the on-board data recorder and are played back in following contacts. The data recorder has two partitions. Data from the spacecraft and scientific instruments are stored in separate partitions, so that scientific operations do not conflict with maintaining the integrity of the spacecraft data. Each partition has a write pointer for recording and a read pointer for play back and behaves like a first-in-first-out (FIFO) memory. When a partition becomes full, the data recorder either overwrites the oldest data or stops recoding according to its setting.

The recorder memory has a total capacity of 8-Gbit. This capacity is three times greater than the amount of data which can be down linked during a ground station contact. Distribution of ground contacts in a day can be irregular. With the large capacity of the data recorder, data rate from telescopes can be determined on a daily basis rather than from the distribution of ground contacts. This feature is well suited for continuous observation in the sun synchronous orbit of *Hinode*. The three telescopes share their partition of the data recorder. Unexpected data volume from one telescope, e.g. by human error

in observation planning or degraded compression efficiency, can result in a loss of data for the other telescopes. To prevent this happening the MDP can be programmed to prevent any of the three telescopes exceeding their allocation.. The MDP monitors the cumulative data recorded by each telescope until a specified limit is reached, at which time it stops further packet addition for that telescope. The assignment of data between the telescopes can be changed on a daily basis.

3.6. Telemetry

Data acquired with the instruments onboard *Hinode* are down linked to Uchinoura Space Center station as well as the Norwegian high latitude (78° 14'N) ground station at Svalbard. Svalbard downlinks every station contact are realized by co-operation with European Space Agency (ESA) and Norwegian Space Centre. Two telemetry channels, S-band (2.2 GHz) and X-band (8.4 GHz), are used. The S-band channel transmits real-time status at 32 kbps. The X-band transmits all of the real-time data and recorded data from the data recorder at 4 Mbps. At the Uchinoura station, the two channels are received simultaneously. At the Svalbard station only recorded data are transmitted via the X-band and no real-time data are available. Note that only real-time data is transmitted via the S-band at the JAXA Ground Network stations for commanding purpose. During the downlink, real time transfer has higher priority than that of recorded data from the DR. Real time data down linked at Uchinoura are sent to ISAS at Sagami-hara, near Tokyo, with the Space Data Transfer Protocol (SDTP) over a TCP/IP network. Recorded data are also sent to ISAS within 90 minutes of the down link. Data taken at the Svalbard station are transmitted to ISAS through the internet, nominally within 90 minutes of their receipt, where it is combined with the data from the Uchinoura station and placed into the ISAS Sirius database.

From the Sirius database, the data are reformatted into FITS files and classified as Level 0 data and are archived on the ISAS DARTS system from where they are made available to the scientific community. The master archive is mirrored to the Solar Data Analysis Center (SDAC) at the Goddard Space Flight Center in Greenbelt and also to a data center in Norway and one at MSSL. The PI institutions in Europe and the United States and several Co-I institutions mirror the data from their instrument to their home institutions.

4. Operations

4.1. Initial Operations and Observations

The month following launch is a period for checking out of the spacecraft and the instruments where the spacecraft and the instruments are operated by their builders. After this period, observation with the scientific instruments starts. The first 90-day observations are planned before the launch and are conducted by the *Hinode* Principal Investigator teams. This provides the instrument teams the opportunity to learn the operational skills needed to run the mission, including the scheduling of operations and archiving data. During this period there are occasional opportunities to access the data archive to retrieve specific data sets. These opportunities enable the user community to test the system and help identify problems before the full data set is released. This is planned to occur about six months after initial operations begin. At that time all the archived data shall be available and all new observations shall be released as soon as after their acquisition.

4.2. Spacecraft Operation

The *Hinode* orbit provides at least two morning and two evening contacts in Japan. Morning contacts provide quick look science data and the evening contacts are used for uploading commands to the spacecraft and science instruments. In addition to the Japanese contacts, the ESA provides 15 contacts per day through Svalbard for downloading scientific data. The average contact time at Svalbard is 11.5 minutes. Allowing 15s for handshaking, approximately 42.5 Gbits of data are downloaded per day.

After the initial period, it is expected that the operation of the spacecraft will become routine. To facilitate safe operation of the spacecraft, patterns of the operations are accumulated and maintained in a knowledge base. In daily operations, a planning tool generates commands for the spacecraft using the knowledge base, predictions of orbital conditions and specification of the downlink stations. The tool also calculates the telemetry allocation for the scientific instruments to be used in planning the scientific program and merges the spacecraft and scientific operations.

4.3. Scientific Operation

Scientific operations are conducted from the ISAS facility located in Sagami-hara, Japan. They are separated into planning and implementation. As shown in Figure 3, the planning process involves monthly, weekly and daily planning meetings. Monthly meetings or teleconferences establish the high level objectives for the next three months and more detailed objectives for the next month. They approve and schedule observing proposals from the external community that were submitted to and approved by the Scientific Schedule Coordinators (SSC). The SSC's are senior scientists designated by the instrument Principal Investigators (PI) who reside at their home institutions. They are responsible for coordinating the monthly observation schedules proposed by the instrument teams with the external proposals. They are also available to assist the external community in preparing proposals and identifying contacts within the instrument teams who can provide proposers with the detailed capabilities of their instruments.

Weekly meetings are held each Friday at ISAS and establish the observing plan, subject to minor changes, for the next week. The plan includes target regions, pointing maneuvers and data recorder allocations. The plan is placed on the *Hinode* operation websites to allow coordination with other observatories. The Daily Meetings are held six mornings a week at ISAS at 10:30 AM local time (01:30 UT) where the daily plan is finalized. In the planning context "days" start at the spacecraft's evening contacts in Japan, which occur at approximately 4:00-7:30 PM local time or 7:00-10:30 UT. At these contacts the instrument commands and observing tables for the next 24 hours are up linked to the spacecraft. With this planning schedule it is possible, in principle, to make minor adjustments to the observing plan as little as eight hours before the observations are made.

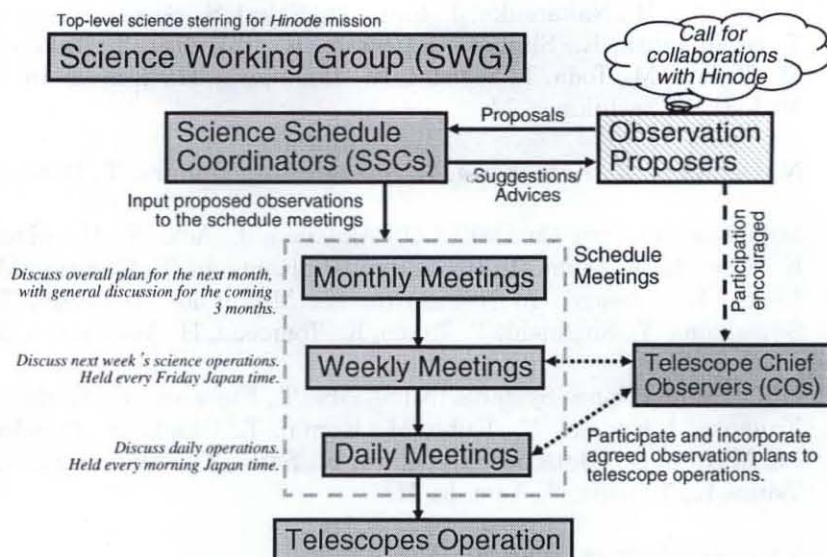


Figure 3 Scientific Operation Planning Flow of Hinode

4.4. Community Involvement

The *Hinode* science teams hope and expect that *Hinode* proves to be a valuable asset to the international scientific community. To expedite collaboration we have created the role of Scientific Schedule Coordinator to provide an interface to the experiment teams. Their role is to both educate proposers as well as to review proposals and schedule observations. Collaboration with other observatories, missions, campaigns, or sub-orbital programs are given high priority. However the data from these observations are also freely available to the community (Matsuzaki et al. 2007).

5. Concluding Remarks

Hinode is a complex satellite that is designed to study primarily how changes in the magnetic field as it emerges through the photosphere affect the higher levels in the atmosphere. It is hoped that the high resolution observations of the vector magnetic field clarify the conditions needed for the onset of magnetic reconnection. The development of the science instruments and objective has been and remains a truly international program and it is hoped that an even broader group of the World's scientists participate in the observations and their analysis.

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